

Table 1. Chemical composition (in wt. %) of materials investigated.

	Al	Cu	Mg	Zn	Fe	Si	Ti	Mn	Zr	Cr	Ni
1933	bal.	0.8-1.2	1.6-2.2	6.2-7.7	≤0.15	≤0.1	≤0.05	≤0.1	0.1-0.18	≤0.05	–
1163	bal.	4.1-4.5	1.3-1.6	≤0.1	≤0.15	≤0.1	0.01-0.07	0.5-0.8	–	–	≤0.05

Table 2. The results of fatigue crack growth duration assessment after bench testing of the RRJ-95 aircraft (No. 95075) wing airframe for different lug sections.

Section No.	Crack propagation direction	Crack length, mm	Number of blocks
1	aft	18.9	1967
2	forward	93.4	9808
3	aft	22.6	5342
4	forward	56.4	5760